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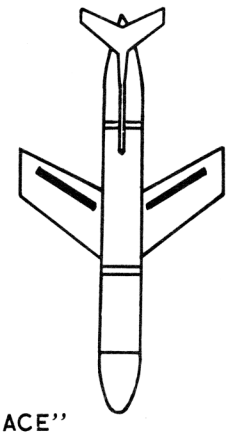
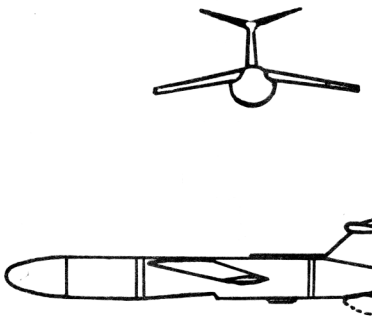
C6 Mace/CGM-13B

Characteristics Summary

TACTICAL MISSILE (RFML) CGM-13B (TM-76A)





"MACE" MARTIN

Wing Area	151.5 sq ft	Length	44.2 ft
Span	22.9 ft	Height	9.7 ft

AVAILABILITY			PROCUREMENT			
Number available			Number to be delivered in fiscal years			
ACTIVE	RESERVE	TOTAL				

STATUS

First YCGM-B (formerly YTM-61B. .	Oct 55	Squadron Deployment	Apr-May 59
First synthetic film flight	Apr 58	Rapid Fire Multiple Launch (RFML) posture proved	May 60
Production	May 58	RFML incorporated tactically . . .	Oct 61
End of R&D	Apr 59		

Navy Equivalent: None Mfr's Model: ---

POWER PLANT

(1) J33-A-41

Allison

THRUST RATINGS

S.L.S.	LB	-	RPM	-	MIN (Inflight)
Max:	5200	-	12,150	-	30
Nor:	4600	-	11,750	-	Cont

BOOSTER

Nr & Model . . (1) * M-16E1-3
Mfr . . . . Thiokol Chem. Corp
Thrust (lb) 101,152
Duration (sec) 2.67

Nominal (70° F)

*With TUX-IS-140 igniter

FEATURES

Tactical Missile capable of destroying targets thru low-level approaches
Swept shoulder type wing
"T" type tail
Self-contained guidance
All weather operation
Low level operation reduces vulnerability to enemy radar detection
Rapid Fire Multiple Launch permits all alert missiles to be launched in approx 10 minutes in salvos of 4
Launched from zero length launcher from hardened sites with assist from booster rocket

Max Fuel Cap: 1029 gal

GUIDANCE

(a) INITIAL: Programmed Pitch and Air-speed Control
(b) MID-COURSE: ATRAN map-matching from lock-on point to end of mission
(c) TERMINAL: None--Missile detonation pro-grammed at end of mission

CONTROL

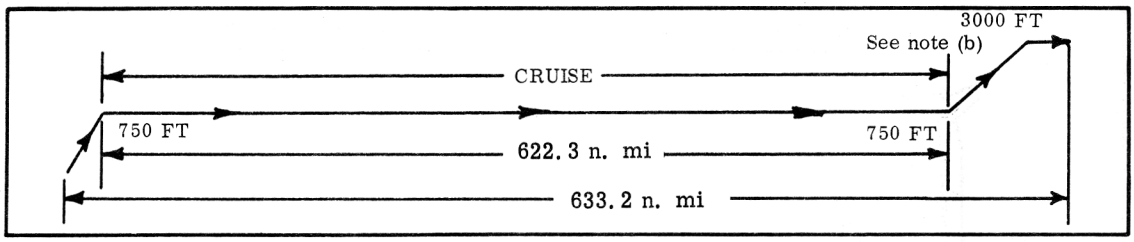
Auto-pilot

DOWNGRADED AT 3 YEAR INTERVALS; DECLASSIFIED AFTER 12 YEARS DOD DIR 5200.10

J. M. Leute 3/31/77

57WC-4983

Characteristics Summary Basic Mission CGM-13B (TM-76A RFML)



PERFORMANCE

ENDURANCE

NOT APPLICABLE

RANGE

633.2 naut mi.
with 1700 lb payload
at 470 knots avg.
in 1.35 hours

SPEED

CRUISE 470 knots at 750 ft alt, power 11,280 RPM (96% RPM)

LAUNCHING

Ground launched from hardened sites on Zero-length launcher. No catapult or runway required but a RATO booster is used for additional thrust.

CLIMB

2200 fpm
sea level, initial climb wt., max power

2110 fpm
750-3000 ft alt, final climb wt., 11,280 RPM (96% RPM)

ALTITUDE

Begin Cruise 750 ft.

End Cruise 750 ft.

LOAD

Payload:	1700 lb
Fuel:	1029 gal
protected	0 %
droppable	0 %
external	0 %

WEIGHTS

Empty	8904 lb
Launch	18,569 lb (see note c)

TARGET ACCURACY

SYSTEM CEP
CEP .70 NM (600 NM)*
CEP 1.20 NM (1200 NM)
*System CEP as an interval estimate with 95% confidence in between .515 NM & 1.658 NM

NOTES

1. PERFORMANCE BASIS:
   (a) Estimated data
   (b) Missile is detonated at altitude at end of mission
   (c) Includes weight of booster rocket
2. REVISION BASIS: To reflect current characteristics and performance, target accuracy redefined, and model designation change.